



PS-28 Cruiser

Q-U-U4AM

CENTRAL WING SECTION S/N:

508

Output control

Weight & Balance record

AIRCRAFT S/N:

C0508

AIRCRAFT SPECIFICATION:

AIRCRAFT TYPE: PS-28 Cruiser
AIRCRAFT S/N: C0508
CENTRAL WING SECTION S/N: 508

INSTALLED EQUIPMENT:

Rotax 912 ULS2 with airbox and thermostats
Woodcomp KLASSIC 170/3/R three-blade ground adjustable propeller
ASI, ALT, VSI, Vertical card magnetic compass, Electric turn coordinator
Electric attitude indicator, Electric directional gyro, Garmin Aera500 GPS
Garmin GTR225A transceiver, PS Engineering PM3000 intercom, King AK451 ELT
Garmin GTX328 transponder, Sandia SAE5-35 altitude encoder, Antennas
Engine RPM indicator, Oil pressure and temperature gauges, CHT indicator
Fuel pressure and quantity indicators, Voltmeter, Engine hours counter
G-205 trim control and PTT on the control sticks, Trims and flaps electrically actuated
Landing light in cowl, Cockpit light, Instrument lighting
AVE-WPST wing tip LED strobe/nav lights
Adjustable pedals, Dual hydraulic brakes, Parking brake
Efficient cabin heating, Carburetor preheating, Wheel fairings tricycle
Upholstery, Paint, Sunshade, Arm supports
BRS LSA softpack parachute

NOTE: Data in blue boxes can be changed.

MEASURING THE AIRCRAFT:

Inflate the tires, drain the fuel from wing tanks.
Level the upper fuselage longeron with a spirit level.
DATUM is on leading edge at RIB # 4.
Measure wheel axle location from DATUM.
Positive arm is behind the DATUM.

LIQUIDS SPECIFICATION:

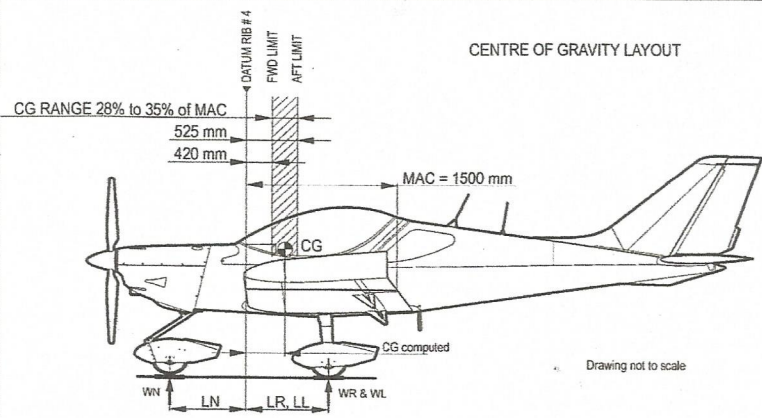
Table with 2 columns: LIQUID TYPE, WEIGHT/VOLUME. Includes FUEL WEIGHT (0.72 KG/LITER), VOLUME (MAX) (114.00 LITRES), OIL WEIGHT (0.90 KG/LITRE), VOLUME (MAX) (3.80 LITRES), COOLANT WEIGHT (1.03 KG/LITER), VOLUME (MAX) (2.50 LITRES).

OPERATING C.G. LIMITS :

Table with 3 columns: C.G. LIMIT, MM, % MAC. Includes MAC (1,500.0), FORWARD CG (420.0), REARWARD CG (525.0), CG RANGE (105.0).

EMPTY WEIGHT C.G. LIMITS :

Table with 3 columns: C.G. LIMIT, MM, % MAC. Includes MAC (1,500.0), FORWARD CG (427.5), REARWARD CG (442.5), CG RANGE (15.0).



AIRCRAFT EMPTY WEIGHT CENTER OF GRAVITY

POSITION OF:

RIGHT MAIN WHEEL (LR)
LEFT MAIN WHEEL (LL)
NOSE WHEEL (LN)
DISTANCE BETWEEN NOSE AND MAIN WHEELS

Table with 2 columns: ARM, MM. Values: 800, 783, -710, 1,502.

WEIGHING POINT:

RIGHT WHEEL (WR)
LEFT WHEEL (WL)
NOSE WHEEL (WN)

Note:

Table with 3 columns: SCALE READING, TARE, NET WEIGHT. Values: 153.2, 152.1, 97.7, 0.0, 0.0, 0.0, 153.2, 152.1, 97.7.

EMPTY WEIGHT: 403.0 KG

MAXIMUM TAKEOFF WEIGHT (CS-LSA / ASTM LSA CATEGORY): 600.0 KG

MAXIMUM USEFUL WEIGHT: 197.0 KG

AIRCRAFT EMPTY WEIGHT C.G. CALCULATION:

MOMENT (KG MM) = WEIGHT (KG) x ARM (MM)

RIGHT MAIN WHEEL
LEFT MAIN WHEEL
NOSE WHEEL

Table with 3 columns: WEIGHT (KG), ARM (MM), MOMENT (KG MM). Values: 153.2, 152.1, 97.7, 800, 783, -710, 122,560.0, 119,094.3, -69,367.0, 403.0, 172,287.3.

AIRCRAFT C.G. = (TOTAL MOMENT / TOTAL WEIGHT) (MM) x (100 / MAC) (%)

RESULT OF AIRCRAFT EMPTY WEIGHT C.G.: 427.5 MM, 28.5% of MAC

DATE: 2014-05-07

ENTITLED PERSON SIGNATURE:

Handwritten signature of Pavel Lukeš



Pavel Lukeš